SUPPLEMENTAL AIRPLANE MAINTENANCE MANUAL

DA 62

STC 62-005

MTOM 2360 kg (5203 lb)

This Supplemental AMM is approved and is valid in conjunction with the basic Airplane Maintenance Manual (AMM).

The limitations and information contained herein either supplement or in case of conflict, override those in the AMM.

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Supplemental Type Certificate	**:**	**STC-62-005**

This document must be carried in the Airplane Maintenance Manual (AMM) at all times. It describes the maintenance procedures.

The technical information contained in this document has been approved under the authority of DOA No. EASA.21J.052.

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CHAPTER 3

GENERAL DESCRIPTION OF THE AIRPLANE

1. Description

The Maximum Take-Off Mass MTOM of the DA 62 is increased to 2360 kg (5203 lb).

This supplemental AMM provides instructions for Continued Airworthiness related to this STC.

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CHAPTER 08
LEVELING AND WEIGHING

Section 08-10
Weighing

2. Weighing With Mechanical Scales under the Wheels

WEIGHING REPORT

Model: DA 62 with STC 62-005 installed Serial Number:_____ Registration:_____

Data with reference to the Type Certificate Data Sheet and the Airplane Flight Manual.

Reference Plane: Vertical plane 2196 mm (86.46 in) in front of the leading edge of wing at the root rib.

Horizontal reference line: Front baggage compartment floor, right side.

Equipment Inventory - dated:_____ Cause for Weighing:_____

Weight and Balance Calculations (Weighing at the wheels)

Weight Condition: Include brake fluid, hydraulic fluid, coolant, engine oil, unusable fuel main tanks (7.57 liter/ 2 US gal) and unusable fuel auxiliary tanks (if installed; 5.0 liter/ 1.32 US gal).

Support	Gross	Tare	Net
MAIN G_{1LH}			
MAIN G_{1RH}			
NOSE G_2			
		Empty Weight	

Lever Arm
X_{1LH} = ... mm (..... in.)
X_{1RH} = ... mm (..... in.)
X_2 = ... mm (..... in.)

Calculate the Empty Weight, G = MAIN G_{1LH} + MAIN G_{1RH} + NOSE G_2.		G =
Calculate the Empty Weight Moment, M = (G_{1LH} * X_{1LH})+(G_{1RH} * X_{1RH}) + (G_2 * X_2).		M =
Calculate the Empty Weight Center-of-Gravity position, X_{CG} = M/G.		X_{CG} =
Maximum permitted all-up-weight: Max. AUW.		2360 kg (5203 lb).
Maximum useful load = Max AUW - G.		

Record the Empty Weight (G) and the Empty-Weight Moment (M) in the Airplane Flight Manual.

Place/Date	Authorizing Stamp	Authorizing Signature

Figure 1: Weighing Report for Mechanical Scales Under the Wheels

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